

Closed-Loop Simulation Study of the Ares I Upper Stage Thrust Vector Control Subsystem for Nominal and Failure Scenarios

*Amy Chicatelli and Chris Fulton
QinetiQ North America, Brook Park, Ohio*

*Joe Connolly and Keith Hunker
Glenn Research Center, Cleveland, Ohio*

NASA STI Program . . . in Profile

Since its founding, NASA has been dedicated to the advancement of aeronautics and space science. The NASA Scientific and Technical Information (STI) program plays a key part in helping NASA maintain this important role.

The NASA STI Program operates under the auspices of the Agency Chief Information Officer. It collects, organizes, provides for archiving, and disseminates NASA's STI. The NASA STI program provides access to the NASA Aeronautics and Space Database and its public interface, the NASA Technical Reports Server, thus providing one of the largest collections of aeronautical and space science STI in the world. Results are published in both non-NASA channels and by NASA in the NASA STI Report Series, which includes the following report types:

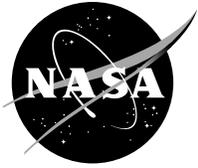
- **TECHNICAL PUBLICATION.** Reports of completed research or a major significant phase of research that present the results of NASA programs and include extensive data or theoretical analysis. Includes compilations of significant scientific and technical data and information deemed to be of continuing reference value. NASA counterpart of peer-reviewed formal professional papers but has less stringent limitations on manuscript length and extent of graphic presentations.
- **TECHNICAL MEMORANDUM.** Scientific and technical findings that are preliminary or of specialized interest, e.g., quick release reports, working papers, and bibliographies that contain minimal annotation. Does not contain extensive analysis.
- **CONTRACTOR REPORT.** Scientific and technical findings by NASA-sponsored contractors and grantees.

- **CONFERENCE PUBLICATION.** Collected papers from scientific and technical conferences, symposia, seminars, or other meetings sponsored or cosponsored by NASA.
- **SPECIAL PUBLICATION.** Scientific, technical, or historical information from NASA programs, projects, and missions, often concerned with subjects having substantial public interest.
- **TECHNICAL TRANSLATION.** English-language translations of foreign scientific and technical material pertinent to NASA's mission.

Specialized services also include creating custom thesauri, building customized databases, organizing and publishing research results.

For more information about the NASA STI program, see the following:

- Access the NASA STI program home page at <http://www.sti.nasa.gov>
- E-mail your question via the Internet to help@sti.nasa.gov
- Fax your question to the NASA STI Help Desk at 443-757-5803
- Telephone the NASA STI Help Desk at 443-757-5802
- Write to:
NASA Center for AeroSpace Information (CASI)
7115 Standard Drive
Hanover, MD 21076-1320



Closed-Loop Simulation Study of the Ares I Upper Stage Thrust Vector Control Subsystem for Nominal and Failure Scenarios

*Amy Chicatelli and Chris Fulton
QinetiQ North America, Brook Park, Ohio*

*Joe Connolly and Keith Hunker
Glenn Research Center, Cleveland, Ohio*

National Aeronautics and
Space Administration

Glenn Research Center
Cleveland, Ohio 44135

Trade names and trademarks are used in this report for identification only. Their usage does not constitute an official endorsement, either expressed or implied, by the National Aeronautics and Space Administration.

Level of Review: This material has been technically reviewed by technical management.

Available from

NASA Center for Aerospace Information
7115 Standard Drive
Hanover, MD 21076-1320

National Technical Information Service
5301 Shawnee Road
Alexandria, VA 22312

Available electronically at <http://gltrs.grc.nasa.gov>

Closed-Loop Simulation Study of the Ares I Upper Stage Thrust Vector Control Subsystem for Nominal and Failure Scenarios

Amy Chicatelli and Chris Fulton
QinetiQ North America
Brook Park, Ohio 44142

Joe Connolly and Keith Hunker
National Aeronautics and Space Administration
Glenn Research Center
Cleveland, Ohio 44135

Abstract

As a replacement to the current Shuttle, the Ares I rocket and Orion crew module are currently under development by the National Aeronautics and Space Administration (NASA). This new launch vehicle is segmented into major elements, one of which is the Upper Stage (US). The US is further broken down into subsystems, one of which is the Thrust Vector Control (TVC) subsystem which gimbals the US rocket nozzle. Nominal and off-nominal simulations for the US TVC subsystem are needed in order to support the development of software used for control systems and diagnostics. In addition, a clear and complete understanding of the effect of off-nominal conditions on the vehicle flight dynamics is desired. To achieve these goals, a simulation of the US TVC subsystem combined with the Ares I vehicle as developed. This closed-loop dynamic model was created using Matlab's Simulink and a modified version of a vehicle simulation, MAVERIC, which is currently used in the Ares I project and was developed by the Marshall Space Flight Center (MSFC). For this report, the effects on the flight trajectory of the Ares I vehicle are investigated after failures are injected into the US TVC subsystem. The comparisons of the off-nominal conditions observed in the US TVC subsystem with those of the Ares I vehicle flight dynamics are of particular interest.

1.0 Introduction

The National Aeronautics and Space Administration (NASA) has been given the mandate to replace the current Shuttle with a new launch system and crew vehicle. The technical and programmatic resources that are needed to achieve this goal are being carried out under NASA's Constellation Program. The new Ares I launch vehicle and the Orion crew module, which rides on top of the Ares I, will take the place of the present Shuttle arrangement. There are primarily two missions that are targeted for the Ares I launch vehicle and Orion crew module. The first is to launch into low Earth orbit (LEO) for meeting with the International Space Station (ISS) and the second is to travel to the Moon.

The Ares I Fault Detection Diagnosis and Response (FDDR) team has been formed to develop the necessary inputs and algorithms that support some of the safety systems that will be built into the Ares I vehicle. NASA document CxP 72244, *Ares I Failure Detection, Notification, and Response (FDNR) System Definition Document (SDD)*, describes in detail the process of notification and mitigation of failure conditions that might occur on the Ares I vehicle. As an element of the Ares I vehicle, the Upper Stage (US) system is divided into subsystems, one of which is the Thrust Vector Control (TVC)

subsystem. This US subsystem is being managed and developed at the NASA Glenn Research Center (GRC), and the GRC FDDR team has been tasked with providing the technical input for the FDNR SDD as it pertains to the US TVC subsystem. Much of this input has been obtained from the US TVC subject matter experts, as well as from using the simulation model of the US TVC subsystem in order to investigate and determine the characteristics of nominal and off-nominal behavior.

Dynamic models and the resulting simulation data provide essential tools that are used for system design, analysis and testing. In particular, when there are no experiment or flight data available, simulation data are necessary to support the development of algorithms for control and diagnostics. Since the dynamic models can be structured so that their outputs represent sensor data streams, they can be used as a virtual test bed in order to simulate the actual hardware system. In addition, the simulation environment provides an opportunity to implement failure scenarios that otherwise might be too costly or dangerous in a hardware test program. As a project matures and hardware test programs are advanced, data become available that can then be used to refine the dynamic models. It is anticipated that this cyclic process can continue as long as the flight vehicle is in service.

In order to support the development of the items described in the preceding paragraph, the capabilities to perform nominal and off-nominal simulations for the US TVC subsystem are needed. In addition, determining the effects on the Ares I vehicle dynamics due to these off-nominal conditions are also desired. Therefore, an integrated simulation model of the US TVC subsystem with the Ares I vehicle simulation code was developed. This closed-loop dynamic model – which incorporated a modified version of the MSFC developed vehicle code, Marshall Aerospace Vehicle Representation in C (MAVERIC)—was created using the Mathworks, Inc.'s Simulink modeling software.

For this report, the effects on the flight trajectory of the Ares I vehicle due to failures in the US TVC subsystem are investigated. These failures include ones that might occur in the actuator components, as well as, those that might possibly occur in the commanded inputs received from the Guidance, Navigation and Control (GN&C) system. Note that this is a limited study that was conducted to support GRC FDDR US TVC goals and was funded by the US TVC Project. In addition, this work was not intended to duplicate or replace the off-nominal MAVERIC simulation studies that were and are continuing to be performed for the US program.

The remainder of the paper is organized as follows. Section 2.0 provides a high-level overview of the Ares I vehicle, with somewhat detailed discussion of the US TVC subsystem. Section 3.0 provides an overview of the integrated simulation model including discussions of the MAVERIC vehicle simulation code, the US TVC subsystem model and the integration process. Section 4.0 describes the nominal and failure scenarios selected for analyses along with the simulation results from those modeled failure scenarios. Lastly, Section 5.0 provides a summary and conclusions.

2.0 Ares I Vehicle Overview

NASA's Constellation Program has described the Ares I/Orion vehicle as replacing most of the Shuttle functions, while being capable of supporting future Moon missions. As such, Ares I has some technologies and components that are based on the heritage of the Shuttle. However, unlike the Shuttle, the crew's exploration vehicle will sit on top of the integrated stack, as shown in Figure 1. In that figure, the Ares I major components are shown: the Orion Crew Exploration Vehicle, the US that includes the J-2X engine and the first stage that includes the 5-segment reusable solid rocket booster. It is not intended to give a complete overview of the Ares I program in this report; rather, a comprehensive report on the Ares I concept and architecture can be found in Reference 1.

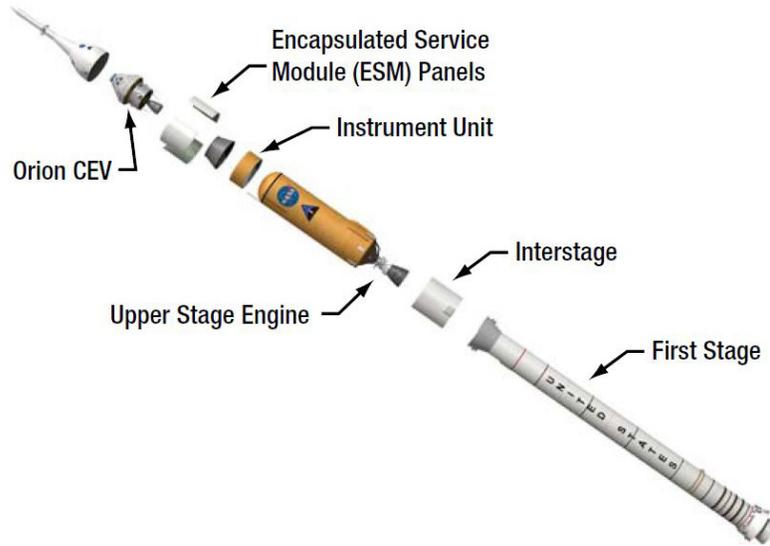


Figure 1.—Major elements of the Ares I vehicle.

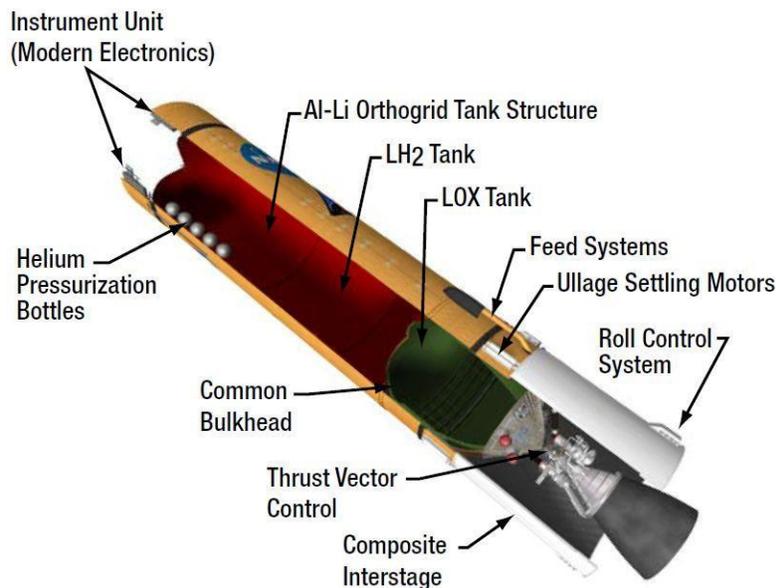


Figure 2.—Ares I Upper Stage.

The following sections briefly describe the US vehicle, the US TVC subsystem and a nominal mission profile.

2.1 Upper Stage System

The US system is used after the First Stage has separated from the Ares I vehicle, and it is required to finish the ascent portion of the flight trajectory. An artist's rendering showing some of the major components of the US is shown in Figure 2.

Basically, the US architecture consists of five main subsystems that are categorized as the following: structures and thermal, main propulsion, roll control and reaction control, thrust vector control and avionics. These subsystems work together to provide the necessary conditions for a successful mission; a detailed report on the Ares I US system can be found in Reference 2. The US program is managed by MSFC along with most of the subsystems. The US TVC subsystem is an exception to this, and it is being developed and managed by the NASA Glenn Research Center in Cleveland, Ohio.

2.2 Upper Stage Thrust Vector Control Subsystem

The primary function of the US TVC subsystem is to control the direction of the J-2X engine thrust in order to maintain the vehicle on its commanded flight trajectory, based on the design requirements listed in Reference 3. Figure 3 illustrates the US TVC subsystem architecture. A description of its operation is contained in the *TVC Operations Concept Document* (Ref. 4). Note that the design is not finalized, but this is representative of the system as it was defined at the time this report was written. Moving across the diagram from right to left, the gaseous hydrogen that is provided by the main propulsion system enters and powers the turbine pump assembly (TPA) via the propellant valves. Once the TPA reaches full power, separate power strings of the hydraulic system transport the hydraulic fluid to the rock and tilt actuators so that they are ready to operate based on commands from the GN&C inputs. In the event of a supply pressure failure, either hydraulic circuit can power both actuators. The actuators are positioned 45° from the vehicle pitch axis and are labeled the rock and tilt actuators, respectively. The commanded stroke of the actuators has been limited to changing the thrust vector by a magnitude of 4° as per the requirements on the US TVC subsystem.

The circulation motor pump shown in Figure 3 is used during pre-launch for thermal conditioning of the US TVC subsystem; it is not used during flight. For this report, pre-launch conditions are assumed to be nominal and have no adverse effects on the ascent phase of the powered flight when the US TVC is active. Hence, a model of the circulation motor pump is neither required for nor included in the system model.

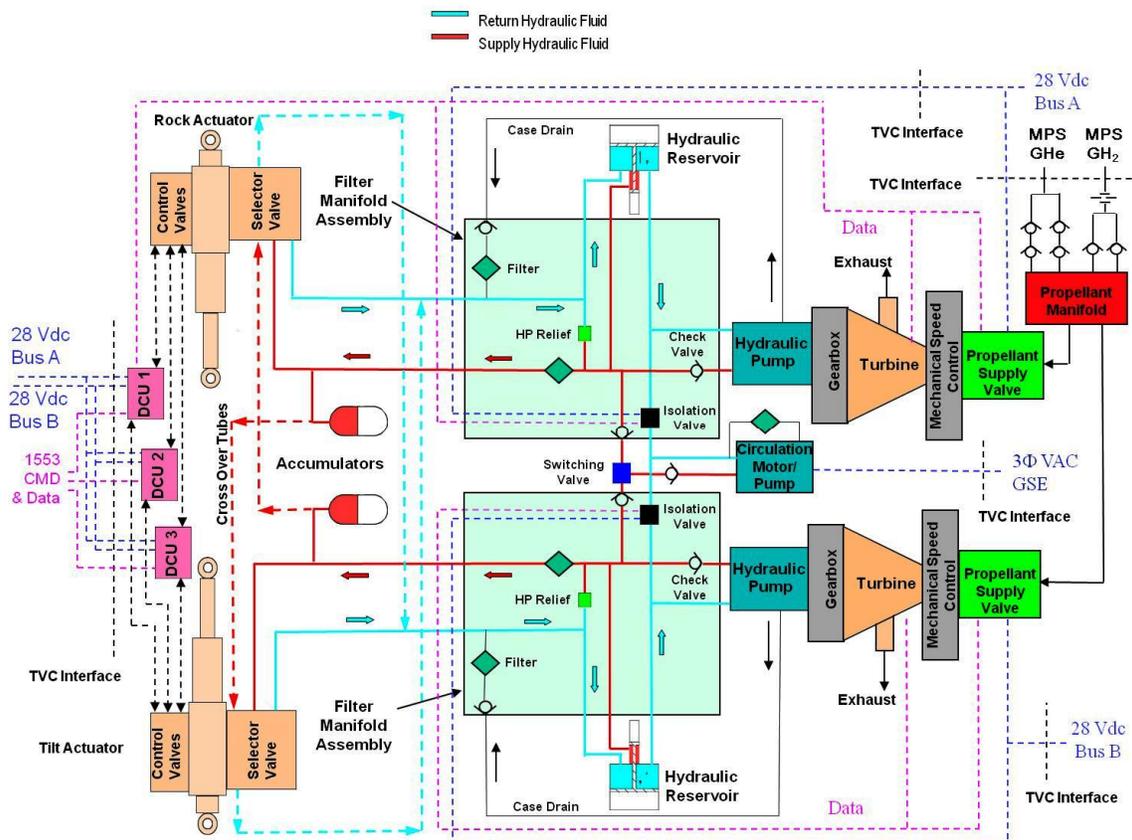


Figure 3.—US TVC subsystem schematic.

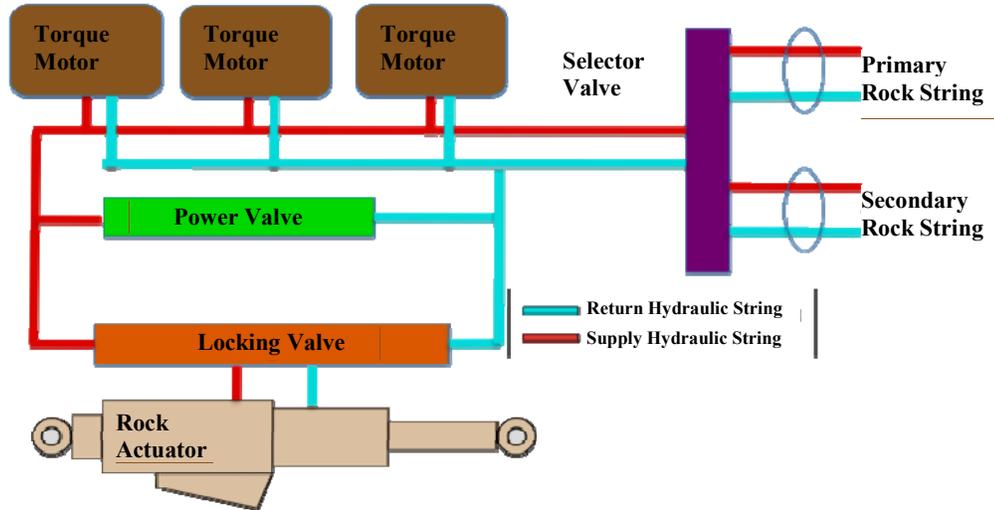


Figure 4.—US TVC actuator assembly.

As expected, the performance of the US TVC subsystem depends on all of its components functioning nominally. In particular, the nominal operation of the rock and tilt actuators is critical to having the US continue on the required flight trajectory that has been defined for its mission. As such, each failure case presented and described in this paper is related to the actuator assembly. Figure 4 shows a simplified schematic of the actuator assembly (reproduced from (Ref. 5)), and a complete description of its operation is provided in Reference 4.

Referring to Figure 4, pressure is supplied from the hydraulic power circuit by either the Primary Rock String or the Secondary Rock String via the selector valve to the three-channel servovalve. Each channel in the servovalve is controlled individually by one of the three torque motors shown above. Downstream from the servovalve, the power valve acts as a summing junction for the three flows out of the servovalve. The actuator is single fault tolerant; a failure in one channel is overcome by the other two channels in a passive voting arrangement. The selector valve is only used in the case of an off-nominal condition. If there is a drop in primary supply pressure, the selector valve moves so that both the rock and tilt actuators are powered by one hydraulic string. Likewise, during flight, the locking valve is only used during an off-nominal condition. If the supply pressure drops too low, the locking valve engages and holds the actuator in its current position. Note that the failure scenarios considered and described in this paper are related to failures in the servovalve or as command errors to the actuator.

2.3 Nominal Mission

The Ares I vehicle is designed for use on LEO and lunar missions. A typical mission profile is shown in Figure 5.

The nominal timeline is being refined as performance requirements and system operation concepts are finalized. Referring to this timeline, for the simulation results presented in this report, the MAVERIC vehicle simulation starts at first stage ignition (launch) and continues until the Main Engine Cutoff (MECO) command. Failure scenarios are focused on the time period that starts when the US TVC subsystem is fully powered (3 sec after Main Engine Start Command) and ends with MECO.

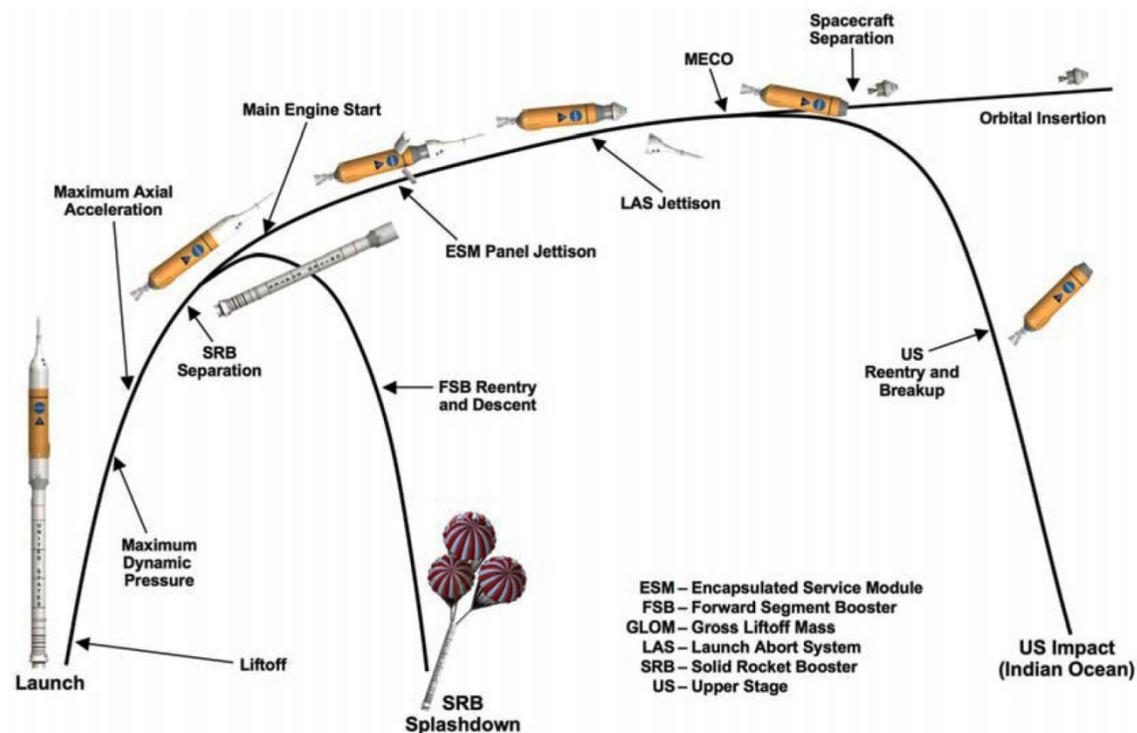


Figure 5.—Ares I flight trajectory.

3.0 Integrated Simulation Model

The integrated Simulink model of the Ares I vehicle with the US TVC subsystem consists of an S-function of MAVERIC (version 3.1) and a dynamic model that represents the US TVC subsystem. The following sections of this report briefly describe the individual simulation models and the process followed to produce the integrated model that was used to perform the closed-loop simulations.

3.1 Upper Stage TVC Simulation

Figure 6 represents the Simulink model of the US TVC subsystem in an open-loop configuration. A complete description of this dynamic model is provided in Reference 5. In addition, simulation results were presented in Reference 5 from hydraulic leak failure scenarios that were computed using this open-loop configuration. In those cases, the rock and tilt commands to the actuators were provided from output data from a stand-alone version of MAVERIC in order to incorporate the GN&C data from the Ares I vehicle. Note that the actuator commands inputs are located inside the actuator blocks, which are not visible in the top-level diagram pictured in Figure 6. Descriptions for the simulation parameters shown in Figure 6 are given in Table 1.

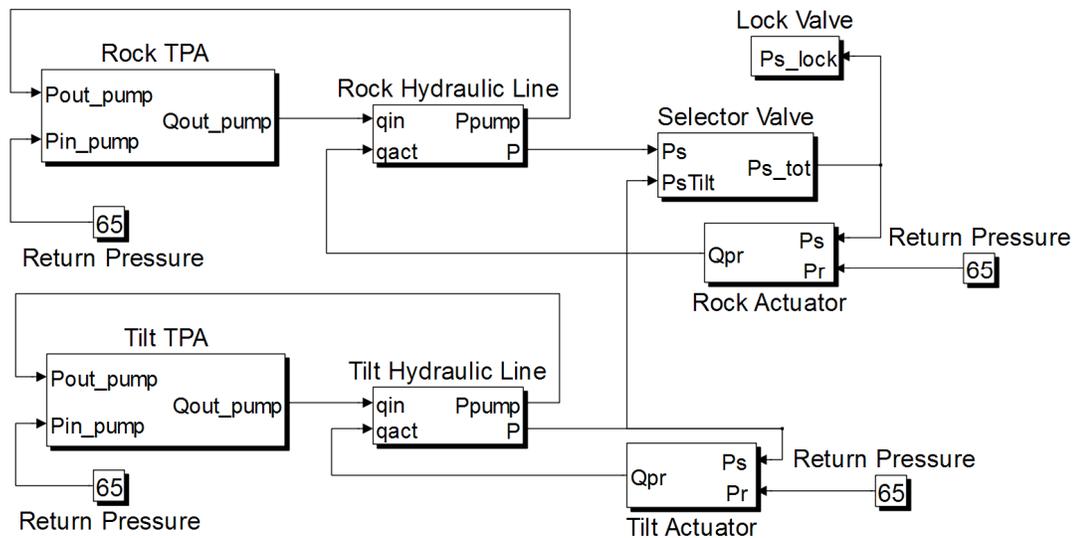


Figure 6.—US TVC Simulink model.

TABLE 1.—DESCRIPTIONS FOR SYMBOLS IN US TVC SIMULINK MODEL

Symbol	Description
Pout_pump	TPA pump discharge pressure
Pin_pump	TPA Pump inlet pressure
Qout_pump	TPA output flow
qin	TPA flow into hydraulic subsystem
qact	Actuator flow into hydraulic subsystem
Qpr	Actuator output flow
P	Hydraulic supply pressure
Ps	Actuator supply pressure
Ps_tot	Selector valve output pressure
Ps_lock	Lock valve supply pressure
Pr	System return pressure

Figure 7 represents rock actuator as it is modeled in the US TVC subsystem simulation model. The command input to the rock actuator component is clearly identified in this lower-level diagram. Descriptions for the symbols shown in Figure 7 are provided in Table 2.

The major components of the US TVC subsystem are modeled with enough fidelity to perform transient analyses. Although in reality the hardware mounted on the thrust cone will not be physically symmetrical, at the time of this report their precise layout is still being determined. Thus for ease of implementation, it is assumed that the rock and tilt components are installed symmetrically, which means that their associated line lengths and connections are the same. Therefore, the system dynamics for the rock and tilt string will be similar.

Each string for the US TVC Simulink model consists of sub-models that include the TPA, hydraulic and actuator subsystems. Comparing the details of the schematic shown in Figure 3 with the block diagram shown in Figure 6, it can be seen that some of the components are not modeled. In particular, the locking valve and the selector valve for the tilt actuator are not included. Under nominal operation during flight, when the US TVC subsystem is active, these components should not be required. Their omission during an off-nominal condition would only be a problem if a failure first occurred in the tilt string, or a failure scenario required their operation. Due to the symmetrical modeling of the system, a failure in the rock string will have a similar effect as a failure in the tilt string.

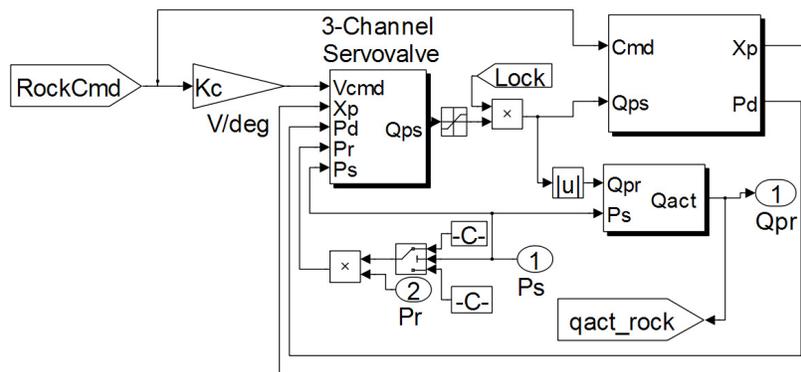


Figure 7.—Rock actuator subsystem Simulink model.

TABLE 2.—DESCRIPTIONS FOR SYMBOLS IN ROCK ACTUATOR SUBSYSTEM SIMULINK MODEL

Symbol	Description
Cmd	GN&C command
Vcmd	Voltage command signal
Xp	Actuator position
Pd	Differential pressure
Pr	Return pressure
Ps	Supply pressure
Qps	Power spool flow
Qact	Actuator flow

3.2 Ares I Vehicle Simulation

The MAVERIC simulation, developed at MSFC, is a software program that allows the modeling of generic aerospace vehicles with the option of three or six degrees-of-freedom. MAVERIC has been used to model and provide simulations for the following applications: Saturn V, a Shuttle-derived cargo vehicle, a Strategic Launch Initiative two-stage-to-orbit launch vehicle concept, an orbital space plane concept and the current crew exploration launch vehicle concepts. A complete description of the MAVERIC software and its use are available in the User's Guide (Ref. 6). The purpose here is not to provide a comprehensive description of MAVERIC and its capabilities. Rather, a brief overview of the simulation is presented with emphasis on its use for this investigation.

For this study, a MAVERIC-based six degree-of-freedom Ares I aerospace vehicle simulation is used. Currently, MAVERIC includes closed-loop ascent guidance, sensor models, inertial navigation and flight control that are specific to the Ares I vehicle. Those aforementioned details are provided in Reference 7. Further, the MAVERIC software program has been used extensively to conduct performance studies for the Ares I vehicle as reported in Reference 8. Hence, the MAVERIC model of the Ares I vehicle is a good candidate for integration with the US TVC subsystem model in order to perform the closed-loop simulation study.

3.3 Integration Process

According to the *TVC Verification and Validation Plan*, an objective of the US TVC project is to have all models and simulations compatible with the MATLAB (Mathworks Inc.) Simulink software package. This necessitated the conversion of the MAVERIC code into a MATLAB S-function so that it could be integrated with the US TVC subsystem dynamic model in Simulink.

The procedure to integrate the MAVERIC simulation and the US TVC subsystem simulation was performed over several steps. The first step involved recompiling the MAVERIC source code into a new stand-alone executable so that simulation results could be compared between the new executable and the results from a validated version of MAVERIC. This was done to ensure that no errors were introduced into the new executable based on platform differences. The recompiling process also required obtaining and compiling the libraries for Global Reference Atmosphere Model (GRAM) and Global Upper Air Climatic Atlas (GUACA), which are required by MAVERIC.

The next set of steps entailed the integration of the MAVERIC simulation into the Simulink environment. This involved the modification of the MAVERIC source code in order to enable the passing of variables between MAVERIC and the US TVC subsystem simulation. At this point, the modified MAVERIC source code was compiled only as object files to be used in the creation of the S-function. These object files in conjunction with the S-function wrapper are linked together to create the MAVERIC S-function. Once the MAVERIC S-function was built, it was used in a Simulink model with no connection between it and an actuator model to ensure that including them in the same Simulink model did not adversely affect the US TVC or MAVERIC models. After that, the actuator model and the MAVERIC S-function were connected together to form a closed-loop system. Integrating only the actuator portion of the US TVC simulation model decreased the complexity of the integration process. Once the integration was completed for this simplified model, the whole US TVC subsystem simulation model was incorporated.

Listed below are the steps that were taken in the integration process. Note that the results from each step were verified using output from the original Ares I MAVERIC simulation.

1. Obtained MAVERIC source code and comparison output results from MSFC.
2. Compiled a new standalone executable code using the following source codes: MAVERIC, GRAM and GUACA.

3. For integration purposes, modified the MAVERIC source code to enable passing of variables between the two simulations and created the MAVERIC S-function. Note that the GRAM and GUACA libraries are also linked in with the MAVERIC S-function.
4. Built a Simulink model that contained the MAVERIC S-function and model of one of the US TVC actuators.
5. Connected the MAVERIC S-function with a single copy of the US TVC actuator model.
6. Connected the MAVERIC S-function with two copies of the US TVC actuator model in order to represent “rock” and “tilt” actuation functionality.
7. Connected the MAVERIC S-function with the complete US TVC subsystem model.

A Simulink block diagram is shown in Figure 8 that illustrates the connections between the two simulation models. At this top-level viewpoint of the simulation, the MAVERIC S-function, labeled “maveric_2Act” in Figure 8, is shown with its GN&C outputs connected as input commands to the US TVC simulation model. There are two parameter settings to the MAVERIC S-function, which select the mission trajectory. The parameters correlate to those used for all of the simulation results presented in this paper, which include the MAVERIC standalone simulation and the integrated closed-loop simulation. The US TVC simulation model, as shown in Figure 6, is represented as the “US TVC Subsystem” block in Figure 8. Therefore, in this closed-loop simulation model the MAVERIC S-function computes the GN&C inputs that are required based on the calculated actuator positions from the US TVC dynamic model. There are other simulation components shown that allow for the following: insertion of actuator command failures at specified simulation times, outputs of simulation results for comparisons and analyses and “scopes” that allow the real-time monitoring of selected simulation variables.

For this simulation, Simulink is configured to use a variable time-step solver. This choice allows for smooth computation through each integration step, minimizing numerical difficulties, which might otherwise cause a fixed time-step solver to crash. In addition, a fixed-step solver cannot be used with the US TVC simulation, because the TPA component simulation, as delivered by a US TVC contractor, uses an S-function that requires a variable-step solver. However, the MAVERIC simulation uses a fixed time-step solver where the time-step is set to a value of 0.01 sec. This could possibly lead to issues when integrated with the US TVC subsystem simulation. For example, if the US TVC simulation was taking large time-steps, the transient response may not be accurate. However, during initial integration of the two simulations, it was seen that when the US TVC subsystem became active, the time-steps used in the solver were on the order of a magnitude smaller than the one used for MAVERIC—generally from 2.0 to 3.5 ms.

The integrated model calculates the new position of the actuators via the following steps:

1. The US TVC Simulink model passes the actuator positions to MAVERIC; this is done through the feedback loop shown in Figure 8.
2. MAVERIC compares the current Simulink timestamp to its own timestamp.
3. If the Simulink timestamp is ahead of MAVERIC, then MAVERIC progresses to “catch up”.
4. The new actuator commands from the MAVERIC GN&C data are passed to the US TVC model via the outputs from the S-function block labeled “maveric_2act” shown in Figure 8.
5. The US TVC model computes one time-step, and the new actuator positions are passed to MAVERIC via the feedback loop. The process is repeated until the end of the simulated mission.

The flow diagram shown in Figure 9 represents the process outlined in the preceding steps.

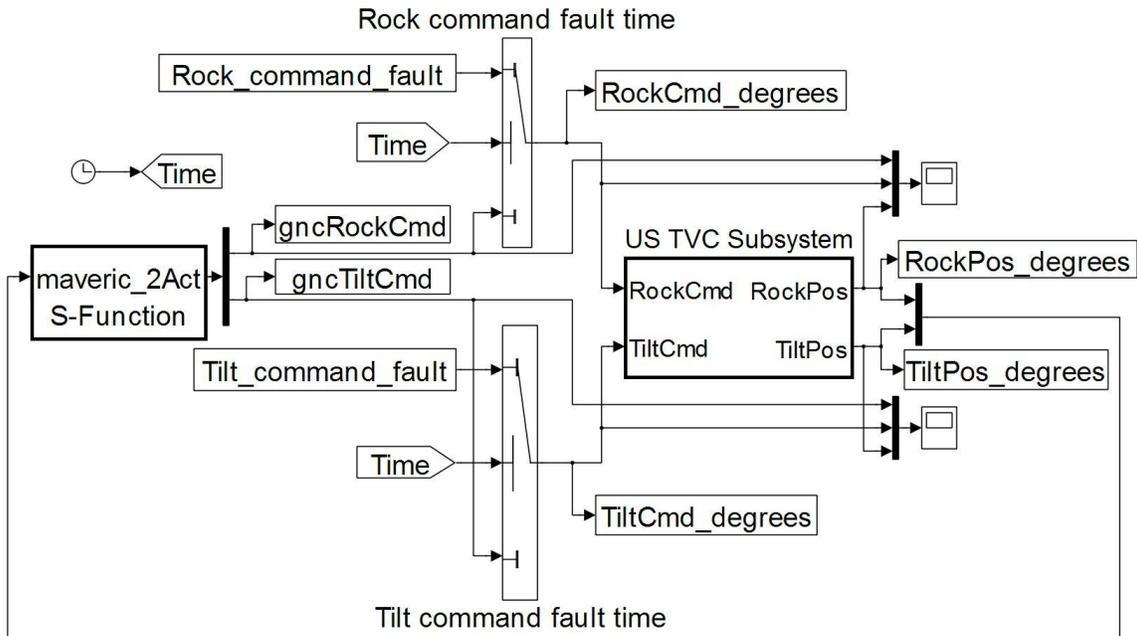


Figure 8.—Simulink integrated model of MAVERIC and US TVC subsystem.

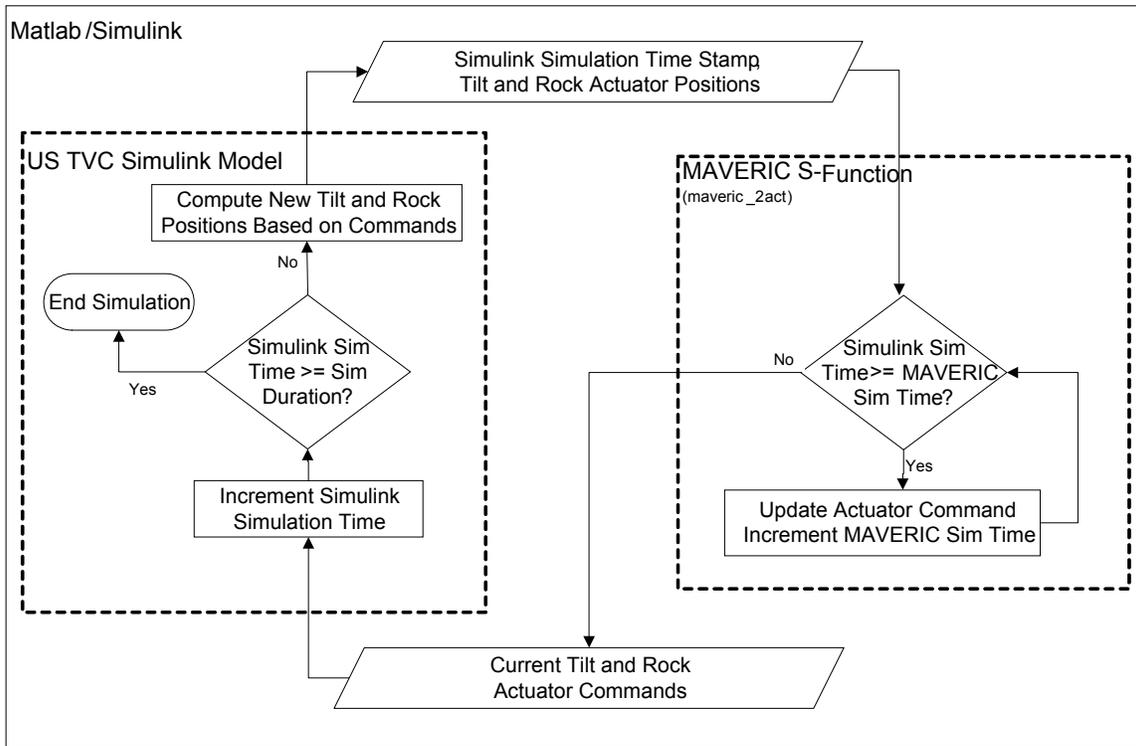


Figure 9.—Integration flow diagram.

As a way to check the reliability of the resulting simulation model, the nominal results from the standalone MAVERIC simulation and the integrated model were compared. Among the comparisons made were plots of vehicle altitude and the gimbal rock and tilt actuator positions. Shown in Figure 10 is a plot of the gimbal rock and tilt positions. The figure shows that the MAVERIC results from the standalone and the integrated models are in agreement for both rock and tilt actuator positions. Therefore, it was deemed the integration of MAVERIC with the US TVC subsystem model was accomplished. Similar analyses were completed for selected failures, and these are presented and discussed in the section for the Failure Scenarios, where again there is agreement for both rock and tilt actuator positions.

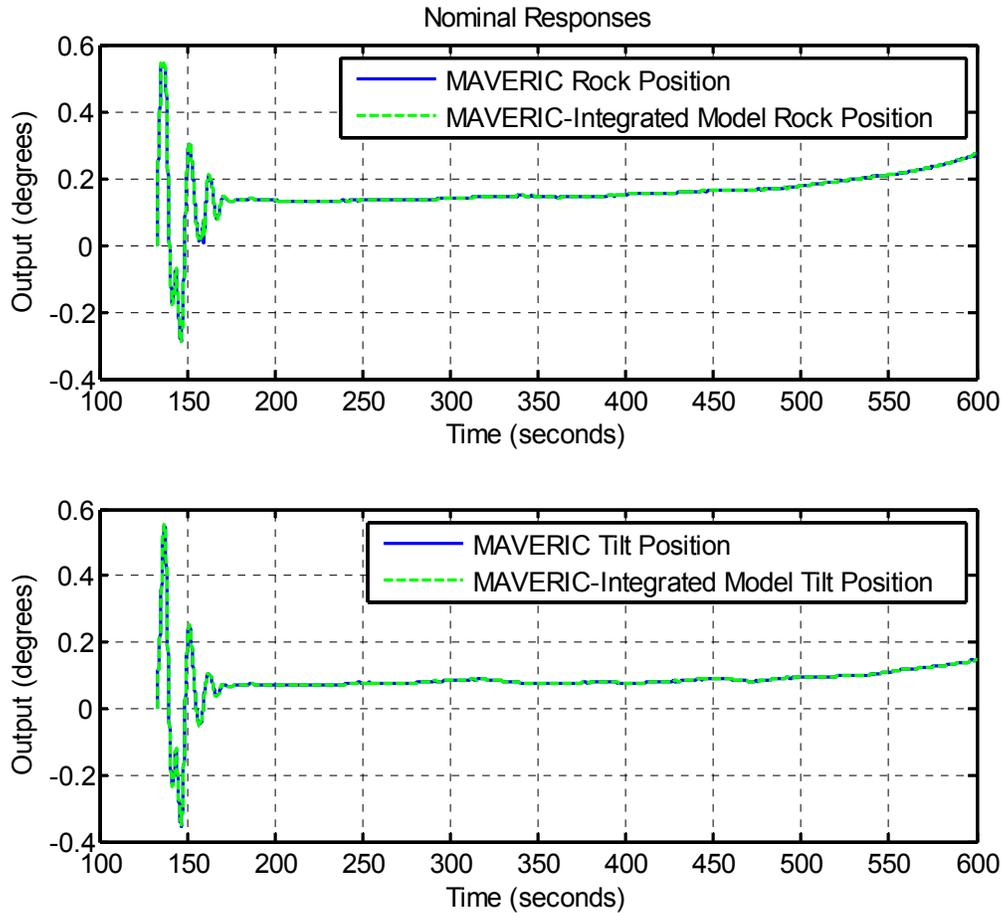


Figure 10.—Comparing MAVERIC output responses for standalone and integrated model.

4.0 Simulation Results for Nominal and Failure Scenarios

The integrated model described in the preceding section was used to generate data under nominal and off-nominal conditions. The fault scenarios selected are those that pose the most risk. In this section, relevant failure scenarios and the nominal scenario are described and the simulation results presented.

4.1 Nominal Scenario

The nominal flight trajectory was selected from among the MAVERIC mission profiles that are included with the version 3.1 software release. The scenario is simulated for 600 sec—the duration of a typical trajectory for the selected mission. For the simulation results presented in this report, the nominal mission is described as a heavy/slow Ares I vehicle that will be launched into LEO for an ISS mission. At the time this work was being conducted, the Shuttle missions were focused on finishing the ISS; so, this was a suitable mission profile. For the integrated model, the MAVERIC S-function and US TVC subsystem model start to communicate with each other at approximately 133 sec as this is the time at which the US TVC subsystem comes on-line. After this time, the positions of the rock and tilt actuators are passed to the MAVERIC S-function and the GN&C commands passed to the US TVC Simulink model, as shown in Figure 9.

Figure 11 shows the nominal responses for the rock command plotted with its respective position and the tilt command plotted with its respective position. The command and position responses are essentially identical, with the response lying on top of the command in Figure 11. The data have been shown from the time when the US TVC subsystem becomes active, at approximately 133 sec, to the J-2X main engine cutoff at 600 sec. It can be observed that the actuator output is always less than the limit of 4°; thus the Ares I requirement for limited actuator movement is met. The GN&C inputs to the rock and tilt actuators have their largest change in positions when the US TVC subsystem is first commanded at 133 sec, and they continue until approximately 175 sec. After that, the position responses of the rock and tilt actuators are relatively steady. Although towards the end of the flight trajectory, from about 350 sec and onward, the position trajectories gradually increase. The flight trajectory selected is not particularly demanding, but it is representative of an Ares I mission profile. Newer versions of the MAVERIC software have flight trajectories that put more demand on the actuation capabilities of the US TVC subsystem. However, for the simulation results presented in this report, the selected mission profile is suitable to illustrate the effects of the selected failure scenarios.

4.2 Failure Scenarios

Using the integrated model that was described in Section 3.0, failure conditions can be inserted in either the rock or tilt strings or both if multiple failures are being investigated. In general, the responses to a particular fault condition, whether implemented in the rock or tilt actuators, are very similar. This is due to the symmetrical modeling of the US TVC subsystem as has been previously noted. Because of this similarity, the fault scenarios presented in this paper are implemented only in the rock string. Additionally, the fault is set to occur at the start of the US TVC operation where the actuators must respond to large commanded inputs. A failure at this time will have a stronger impact on the system response; because, it occurs during a critical time for the performance of the US TVC subsystem. Table 3 lists the US TVC failure scenarios investigated for this paper.

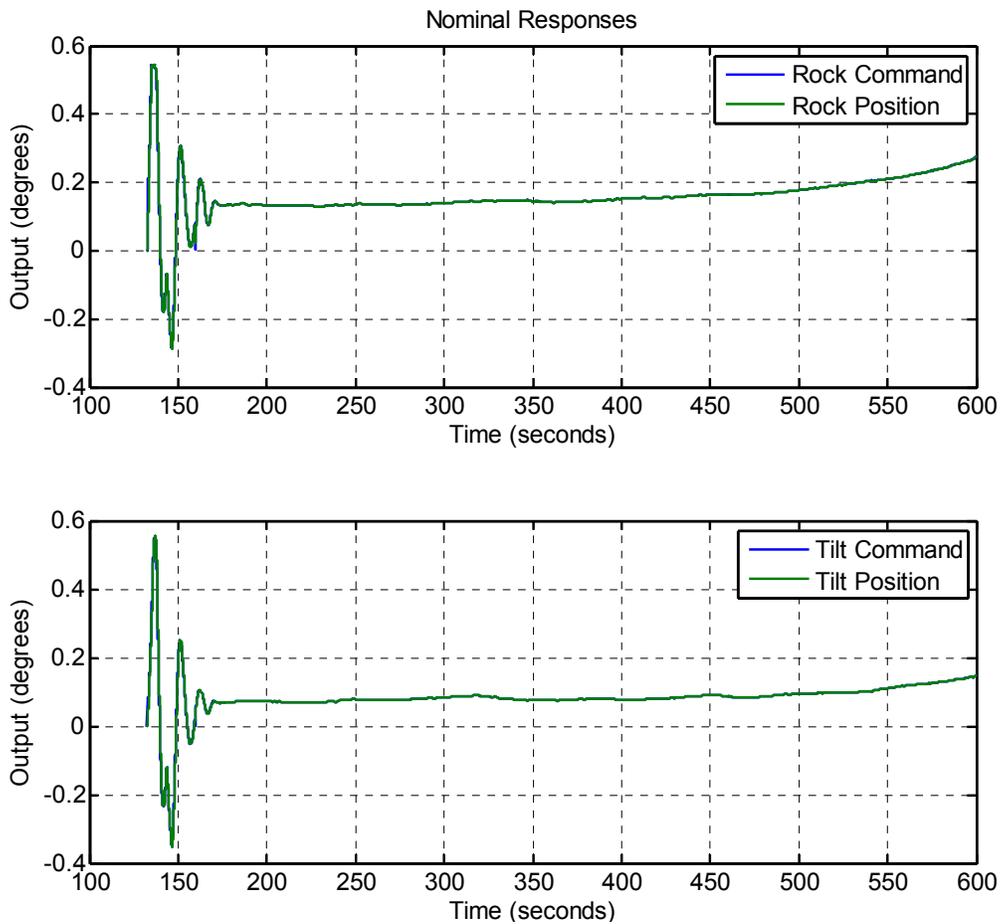


Figure 11.—Output responses for the nominal scenario.

TABLE 3.—US TVC FAILURE SCENARIOS (FS) INVESTIGATED

No.	Description
1	Hard over failure in 1-channel of the rock actuator servovalve
2	Hard over failure in 2-channels of the rock actuator servovalve
3	Zero current failure in 1-channel of the rock actuator servovalve
4	Zero current failure in 2-channels of the rock actuator servovalve
5	Zero current failure in 3-channels of the rock actuator servovalve
6	Commanded hard over to rock actuator (4° input)
7	Commanded null position to rock actuator (0° input)

Next, the results and discussion are presented for each of these failure scenarios. Note that the results are shown for the time period of interest for operation of the US TVC system. Namely, the plotted simulation results start at approximately 132 sec and continue until the nominal mission ends at 600 sec or until the inserted failure causes an end to the mission trajectory. Also, the simulation results for the failure scenarios are designated by FS, which can be found in each figure that compares the nominal and failure scenario results.

These failure scenarios do not take into account any recovery strategies that would most likely be in place as a result of fault detection, isolation and recovery (FDIR) algorithms for abort conditions that would be designed for the Ares I vehicle. Therefore, when a failure occurs, the integrated simulation continues to run with no mitigating actions taken. Therefore, results may occur from the failure insertions that would not otherwise be expected.

4.2.1 FS No. 1—Hard Over Failure in 1-Channel of the Rock Actuator Servovalve at $t = 133$ sec

A hard over failure condition exists when an actuator is fully extended or fully contracted and does not respond to commands sent from GN&C. In other words, the US TVC does not have the capability to keep the US on its mission trajectory. As stated previously, the servovalve has three inputs (i.e., channels) that make up the pilot flow in each actuator; a failure in one of them has only a minor adverse effect on the ability of that particular actuator to function normally. This is because, the other two nominal channels of the servovalve can compensate for the failure in the third one. The purpose of this failure case is to verify this effect. The failure scenario is implemented in one channel of the rock actuator servovalve, which is shown in Figure 4. As expected, the US TVC subsystem is able to control the US and maintain the mission trajectory. Therefore, the responses for this failure scenario look very similar to the nominal ones. The impact of the failure can be seen in Figure 12 where the rock command is off-set to compensate for the one failed servovalve in the rock actuator. For these comparisons, the output responses are close to the nominal ones, because the GN&C system keeps modifying the rock actuator command until the desired trajectory is achieved. The results show that, with this failure, the nominal mission will be completed successfully due to the fact that the rock actuator assembly can be operated with one failed channel.

4.2.2 FS No. 2—Hard Over Failure in 2-Channels of the Rock Actuator Servovalve at $t = 133$ sec

In order to have the rock actuator go into a hard over condition, there will need to be at least two faults in order to violate the single-fault tolerant design. Since, the US TVC cannot operate in a nominal manner with an actuator in a hard over position; it will not have the capability to keep the US on its mission trajectory under this failure condition. The objective of this failure scenario is to show that result, and it is implemented by having two channels of the servovalve fail so that the rock actuator is held in the fully extended position. In this case, the failure impacts the nominal trajectory in a severe manner. The US cannot be controlled using only the properly functioning tilt actuator. The altitude output responses shown in Figure 13 for the nominal and failure scenario do not start to diverge until about 150 sec. While this is a subjective measurement, it is adequate for the purpose of comparing the effect on the vehicle trajectory due to the hard over failure. Note that the position of the actuator has been limited to a magnitude of four degrees as per the requirements on the US TVC subsystem and shown in Figure 14. As the rock actuator moves to its maximum position, the rock command from the GN&C system compensates by sending inputs to move a -4° . Correspondingly, the GN&C inputs to the tilt actuator try to compensate for the rock actuator—the result of being 45° off axis—driving that actuator to its limit as well. The abrupt change in the position of the tilt actuator, as seen in Figure 14, could be due to the GN&C function reacting to the vehicle's response to the hard over condition.

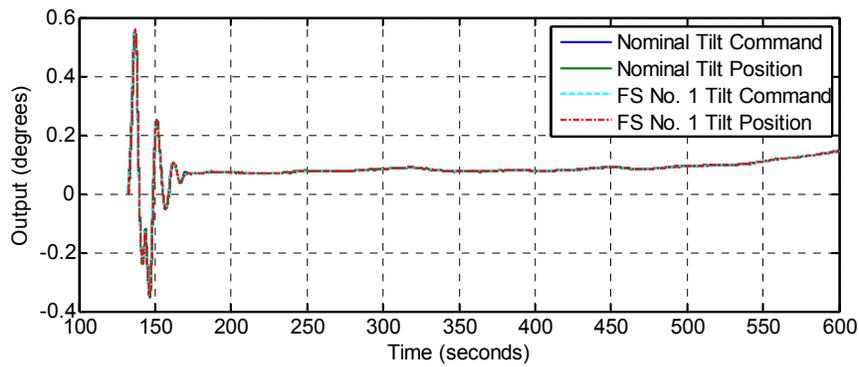
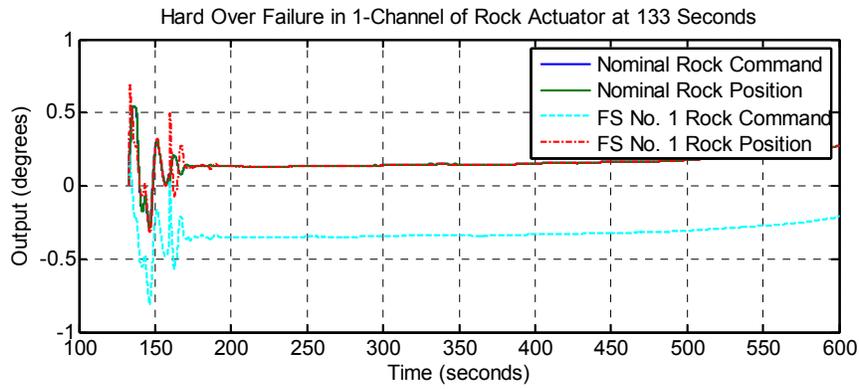


Figure 12.—Comparing output responses for nominal and failure scenario (FS) one.

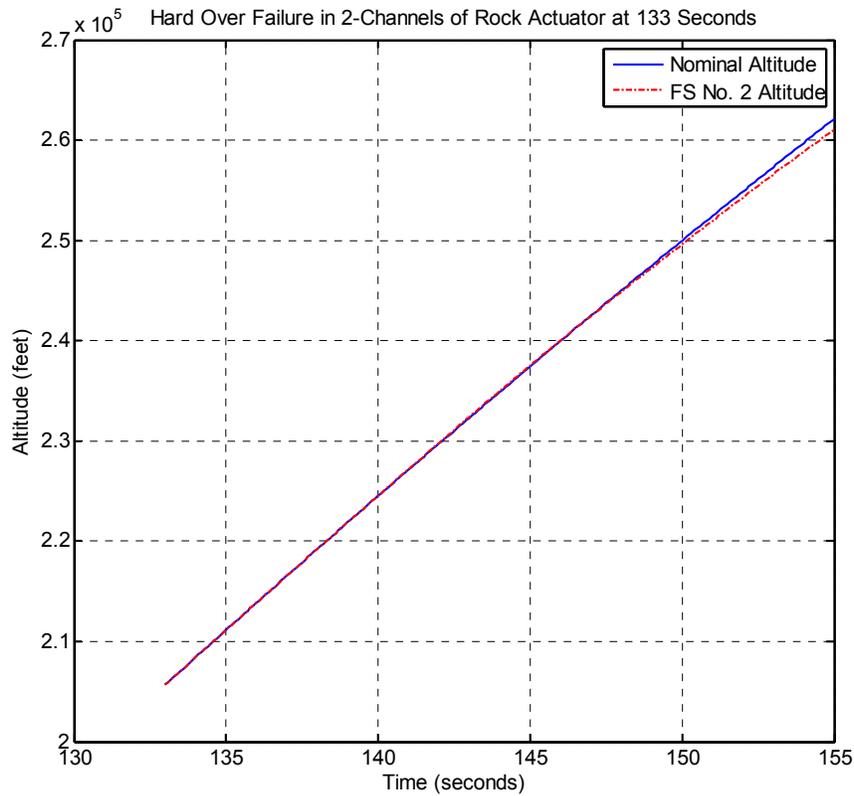


Figure 13.—Comparing altitude responses for nominal and failure scenario two.

Figure 14 illustrates the sequence of events that occur. Looking at the tilt position, the impact from the hard over failure in the rock actuator is evident at around 138 sec, which is when the nominal and failure trajectories start to significantly deviate. This is still a qualitative measurement of when the fault would manifest itself in the sensor data, because the tilt position is still in the range of nominal behavior. When the delta in the altitude responses between the nominal and failure scenario is plotted, the deviation between the two is also seen to start at approximately 138 sec. From a performance point of view, this simulation shows that the Ares I vehicle will not meet its trajectory target with two servovalve failures that result in a hard over condition in the US TVC actuator subsystem. In this case, the integrated simulation has a numerical instability and stops at approximately 170 sec.

Previously in Figure 10, it was shown under nominal conditions the MAVERIC-integrated model performed the same as the standalone simulation. Similarly, the performance of the integrated model under a failure condition was also investigated in order to demonstrate a successful integration process.

The process for inserting the hard over failure into the MAVERIC standalone simulation was constrained by the rules imposed by the code and was different than for the integrated model. Despite these differences, the fault input was reasonably replicated. In the first plot of Figure 15, the rock positions illustrate the effect from the hard over failure where both positions are driven to 4°, which is a hard over condition. In the second plot of the same figure, minor differences in the tilt positions can be seen at approximately 145 sec where the peaks in the responses are slightly offset. Concerns about discontinuities in the tilt position response from the integrated model are also in the tilt position response from the standalone simulation. Therefore, these discontinuities are not due to the integration process.

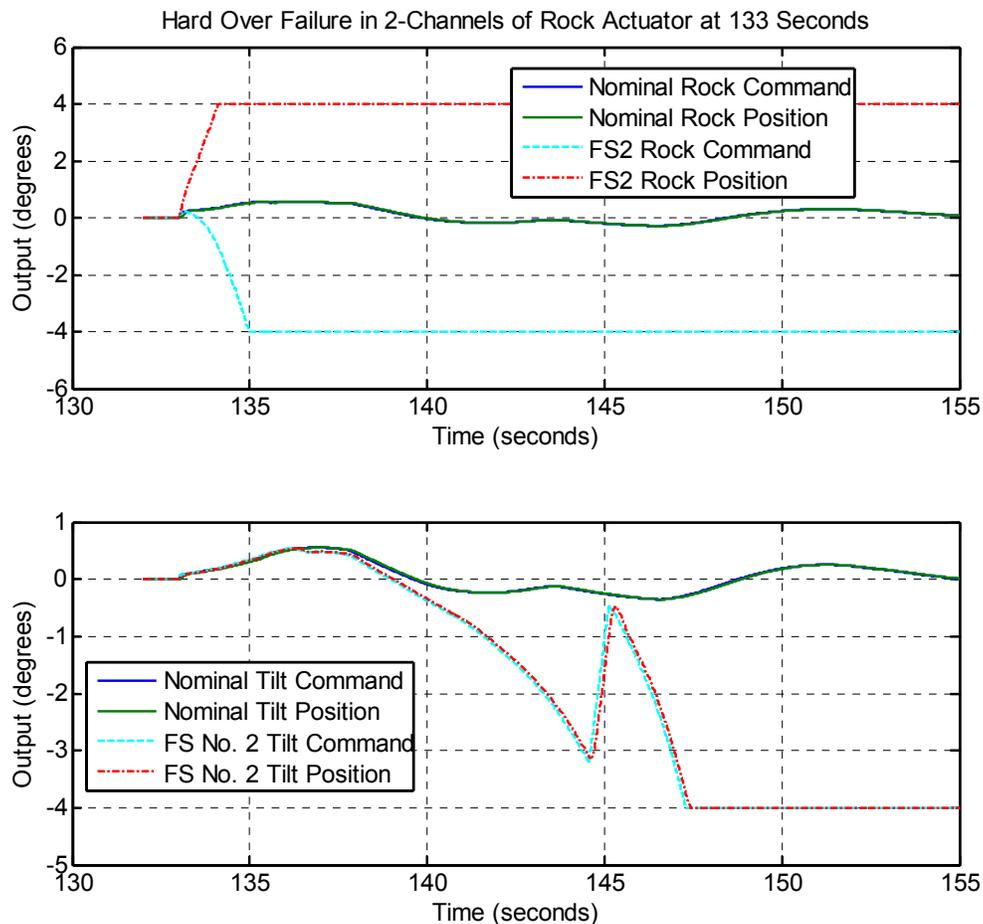


Figure 14.—Comparing output responses for nominal and failure scenario two.

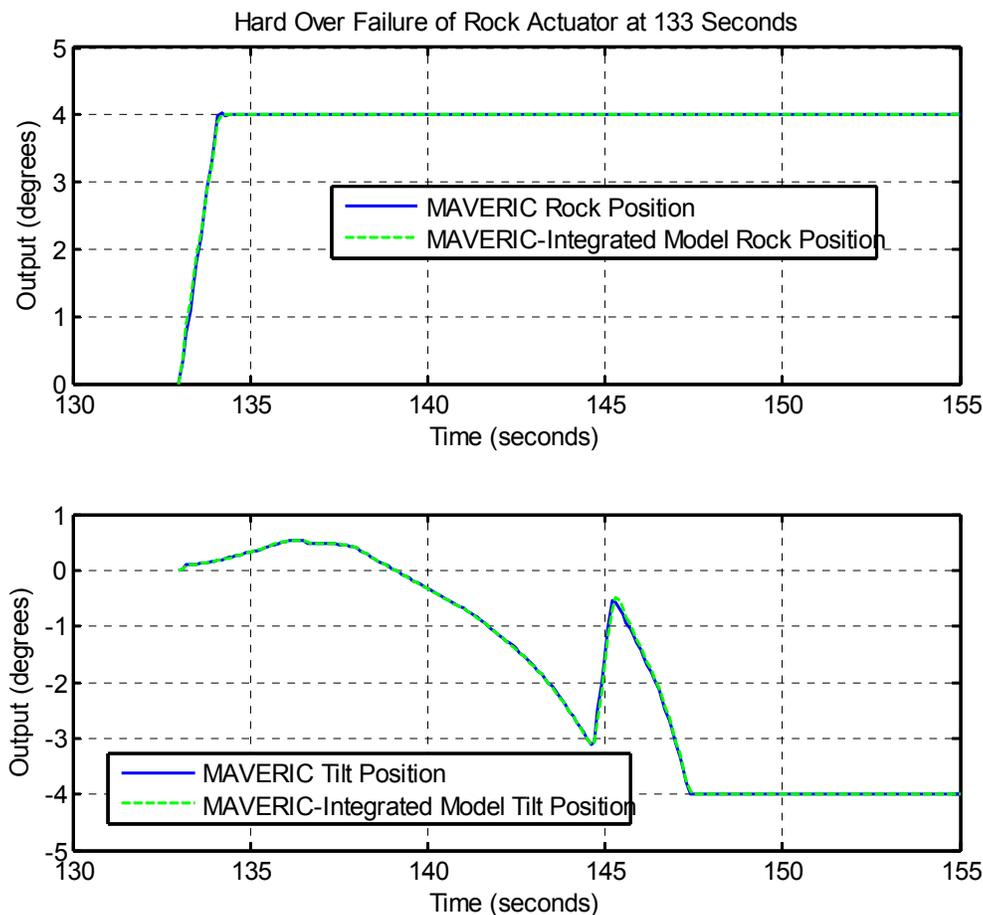


Figure 15.—Comparing MAVERIC output responses for standalone and integrated model.

4.2.3 FS No. 3—Zero Current Failure in 1-Channel of the Rock Actuator Servovalve at $t = 133$ sec

When the actuator is moving, each torque motor contributes to the overall movement of its associated actuator. If there is no current through the torque motor, there is no contribution from that particular channel. The purpose of this failure scenario is to confirm that the actuator can still nominally perform its function with the loss of one torque motor. This failure scenario is implemented in one channel of the rock actuator servovalve. As expected, losing one of the torque motors (i.e., no current) in the servovalve has no adverse impact on the ability of the Ares I vehicle to meet its mission trajectory. In general, the output responses for the failure scenario are very similar to the nominal ones for the variables that have been previously shown. Consequently, they are not shown here. Because the responses are similar, the fact that a failure has occurred in the rock actuator servovalve will not be readily observed in the sensor data for the rock and tilt position measurements. However, there is a sensor embedded in the actuator assembly that monitors the torque motor current, and that signal can be used to provide indication that there is a problem with the torque motor since torque motor current should not be zero when the associated actuator is moving. Although the failure has not caused a severe impact to the Ares I mission, awareness of the loss of redundancy during flight is desirable both for onboard operation and post-flight analyses.

4.2.4 FS No. 4—Zero Current Failure in 2-Channels of the Rock Actuator Servovalve at $t = 133$ sec

This failure scenario is similar to the previous one, except that now there are two channels in the rock actuator servovalve with no current through them. The overall simulation results are not substantially different from the previous failure scenario which, in turn, is similar to the nominal case. Using similar reasons as in the preceding example, indication of this fault condition would need to be observed in something other than the actuator position measurement. As before, although the simulation results do not indicate a severe impact to the Ares I mission, it would still be beneficial to know that there has been a failure in the rock actuator subsystem.

4.2.5 FS No. 5—Zero Current Failure in 3-Channels of the Rock Actuator Servovalve at $t = 133$ sec

If each channel of the actuator servovalve has a failed torque motor, the actuator will not move. The results from this study indicate that at least one working torque motor is required for the actuator to operate. The goal of this failure scenario is to demonstrate that the actuator cannot nominally perform its function with the loss of all three torque motors, and this failure scenario is set up with all three channels of the rock actuator servovalve concurrently having a zero current failure. Evidence of the diversion between the altitude responses between the nominal and failure scenario occurs at approximately 160 sec, as seen in Figure 16. The first plot of Figure 17 clearly indicates an off-nominal situation at the time of the insertion of the fault, because the rock actuator is not moving. In spite of the fact that the rock actuator is nonresponsive, using only the tilt actuator, the controller is able to maintain the desired trajectory up to about 160 sec. At this point, the controller can no longer maintain the desired trajectory using only the tilt actuator. For most of the trajectory, the responses for the tilt actuator command and position are well within the maximum allowable range, but their actual magnitude and transient responses are very different from the nominal ones. The error between the nominal and failure scenario is obvious in the second plot of Figure 17. Eventually, the tilt actuator position goes into a hard over condition, which is indicated in Figure 17 at approximately 325 sec.

If only the allowable limits on the actuator position are being used, these results show that those measurements may not clearly indicate that there is an off-nominal condition. However, using the error between the commanded position and the actual position will provide an earlier indication that there is a problem; because, the rock actuator is not responding to inputs from GN&C. An earlier indication is desired, because it provides more time for remediation of the failure. In addition, as noted before, using the measurements for the torque motor current sensors will also provide an earlier indication that there is a problem. Since, under nominal conditions the rock actuator should be responding, which means that the torque motors should have measurable currents. In this failure scenario, the currents are zero, and it can be inferred that there is a problem.

As done before with the hard over failure, a comparison between the MAVERIC standalone simulation and the integrated model under a null condition failure (i.e., zero current) was also investigated in order to demonstrate a successful integration process.

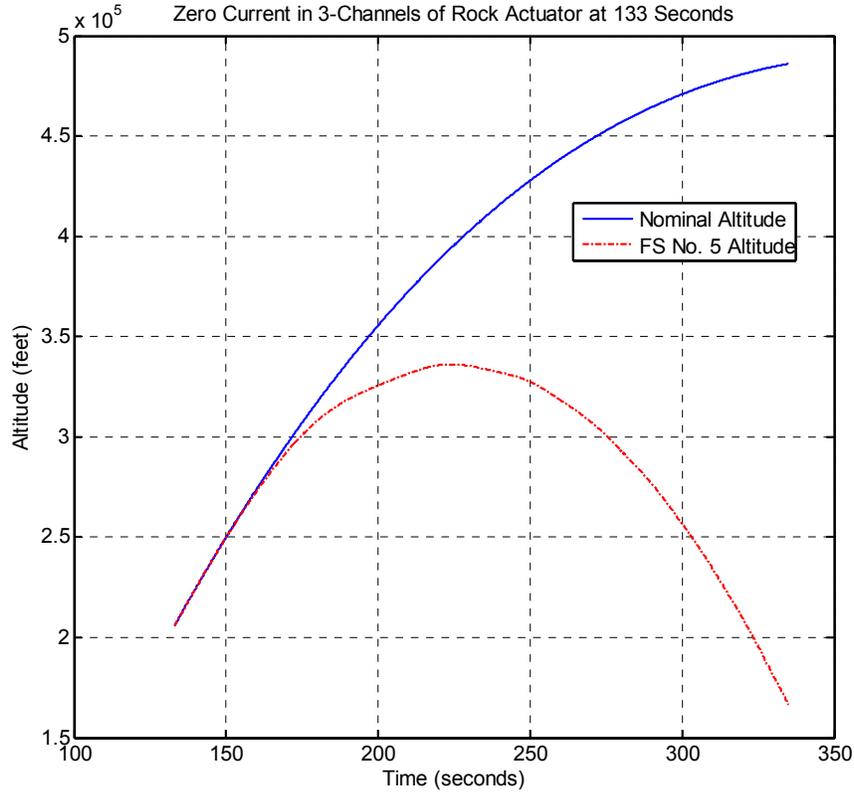


Figure 16.—Comparing altitude responses for nominal and failure scenario five.

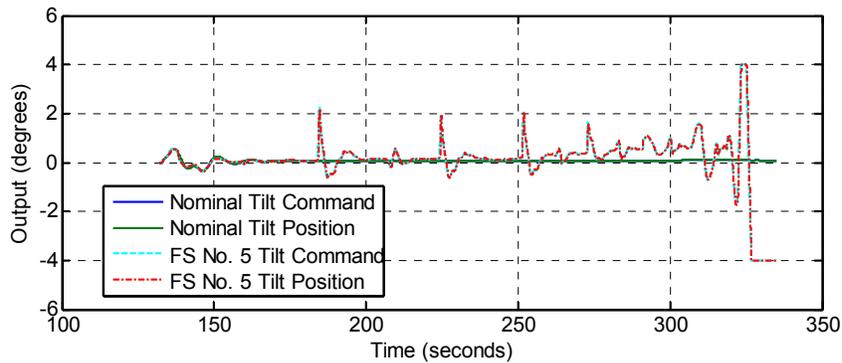
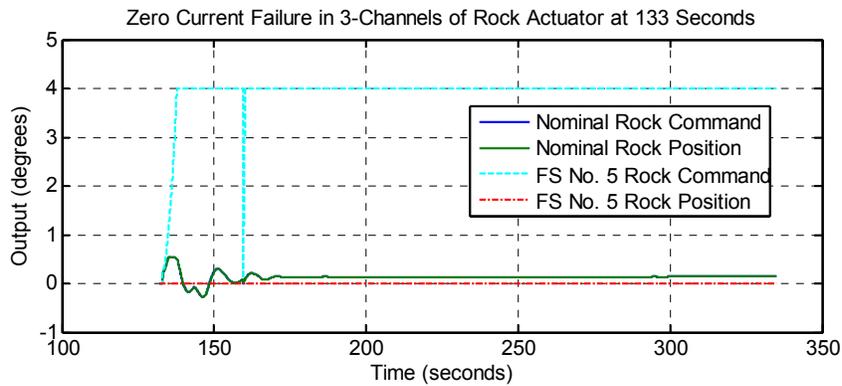


Figure 17.—Comparing output responses for nominal and failure scenario five.

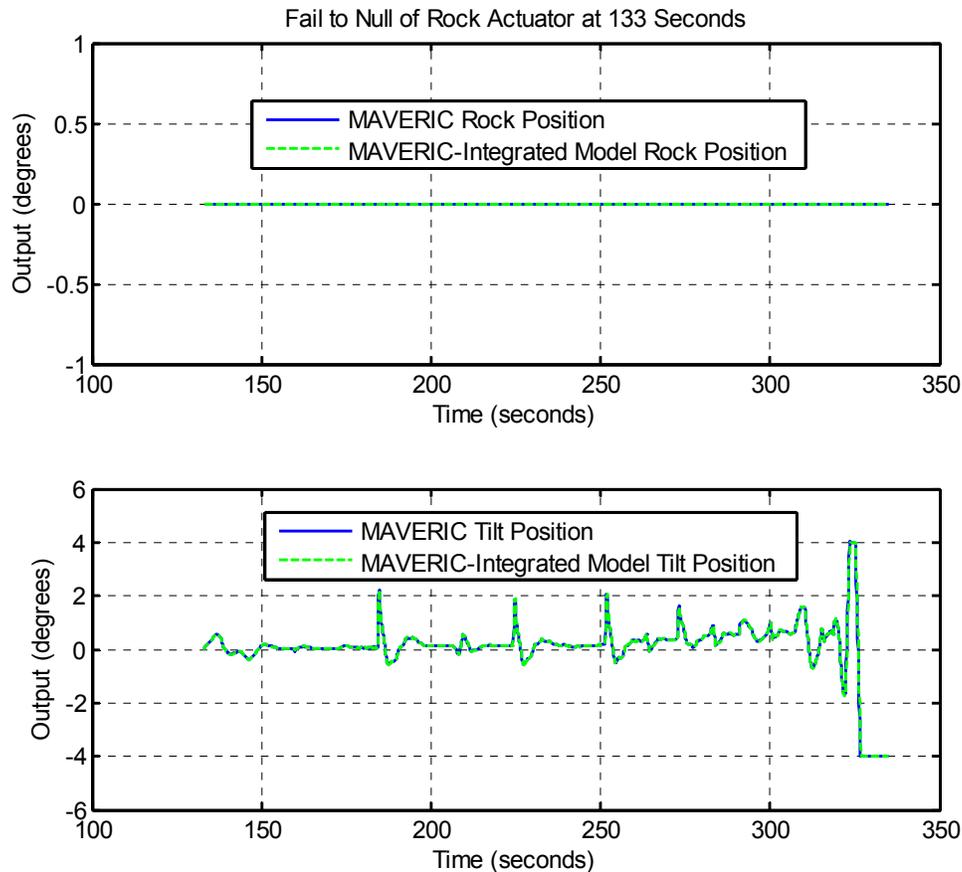


Figure 18.—Comparing MAVERIC output responses for standalone and integrated model.

The null failure was inserted into the MAVERIC standalone simulation in the same manner as was for the hard over scenario. However, in this case the failure is easily repeated because for the null condition the actuator remains at rest. In Figure 18, the output responses show good agreement, and once again there were no problems indicated with the integration process for the failure scenario investigated.

4.2.6 FS No. 6—Commanded Hard Over to Rock Actuator at $t = 133$ sec

As noted in previous failure scenarios, a hard over condition is the result of the actuator going into a fully extended or fully contracted position. As shown in the second failure scenario, this can be the result of a hardware failure. In addition, it can also be the result of an error in the input command to the actuator for instance, with a software error being one possible cause. The reason for this failure scenario is to demonstrate the effects of a command input error; for this example, an erroneous command of 4° is sent to the rock actuator that results in a hard over condition. This failure scenario was also included to illustrate the functionality of the integrated model, and its ability to model failures in the command inputs. As presented in Figure 19, the effect on the trajectory of the altitude response is similar to the second failure scenario where there was a double failure in the servovalve subsystem of the rock actuator assembly, which resulted in a hard over condition. Looking at the first plot in Figure 20, the rock position moves to its maximum value of 4° as commanded. As a result, the commands sent to the tilt actuator try to compensate for the rock actuator where eventually the tilt actuator is driven to its limit as well. As illustrated by the off-nominal response in the altitude trajectory, the US cannot be controlled with just the tilt actuator.

It is important to note that failures resulting from different causes can have similar effects on the system. Arès I will require diagnostic systems to detect the root cause of these failures and to isolate the failure to a specific component or subsystem.

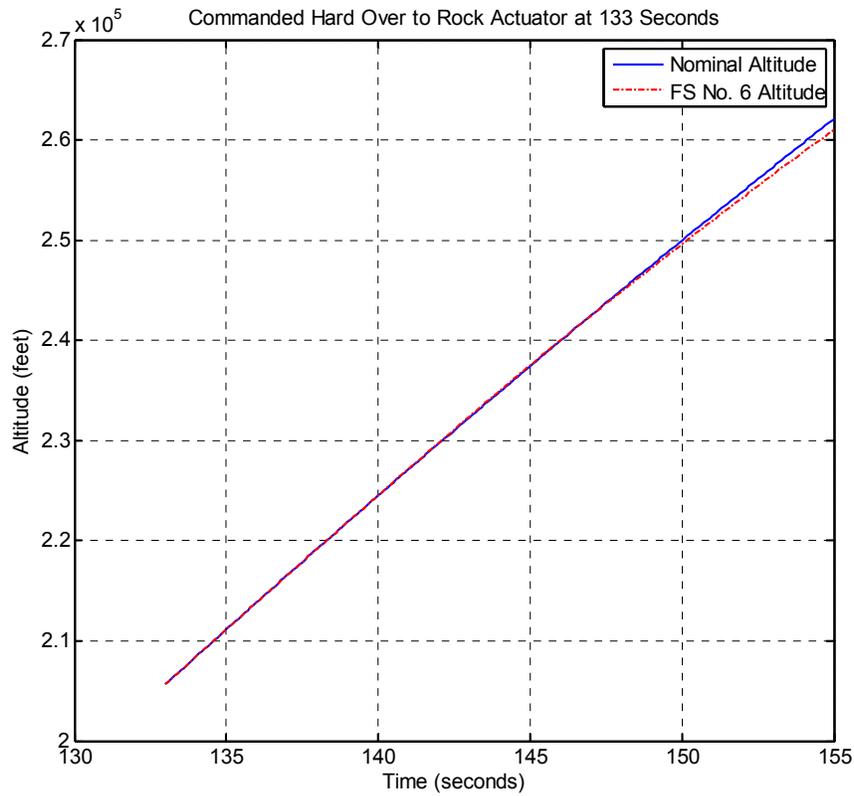


Figure 19.—Comparing altitude responses for nominal and failure scenario six.

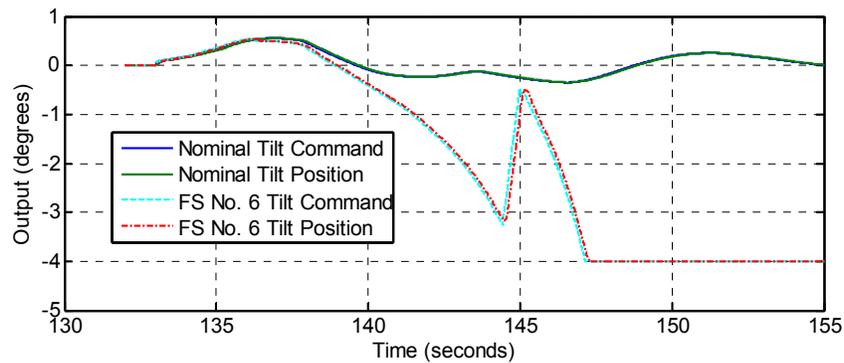
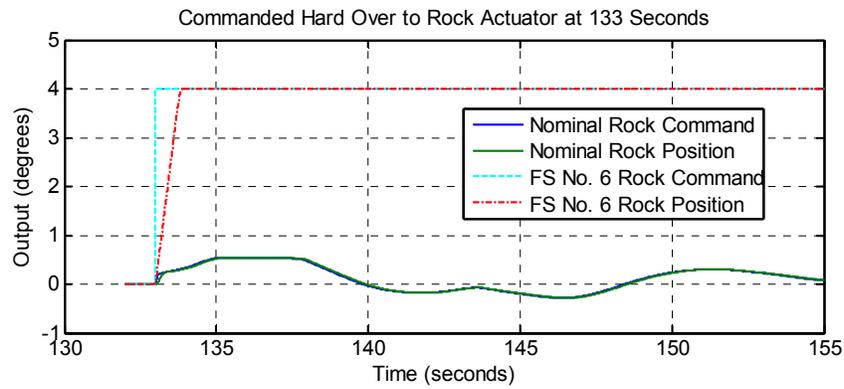


Figure 20.—Comparing output responses for nominal and failure scenario six.

4.2.7 FS No. 7—Commanded Null Position to Rock Actuator at $t = 133$ sec

This failure scenario is also the implementation of a command input error. The motivation for this case is to have an erroneous command that places the rock actuator at the null position, which is at 0° . In other words, there are no commanded deflections in the rock actuator position. The fault insertion occurs at 133 sec—the beginning of US TVC operation. Divergence in the altitude response between the nominal and failure scenario occurs at approximately 160 sec, as seen in Figure 21. In the first plot of Figure 22, the rock actuator position clearly indicates an off-nominal situation at time of the insertion of the fault. While the tilt actuator position is within its nominal range of 4° , the error between the nominal and failure scenario is obvious in the second plot of Figure 22. Eventually, the tilt actuator position goes into a hard over condition at approximately 325 sec. Note that the response for this failure is similar to that for failure scenario five. So once again, it can be seen that it becomes important to use information other than that provided from the trajectory error to properly isolate the cause of failure.

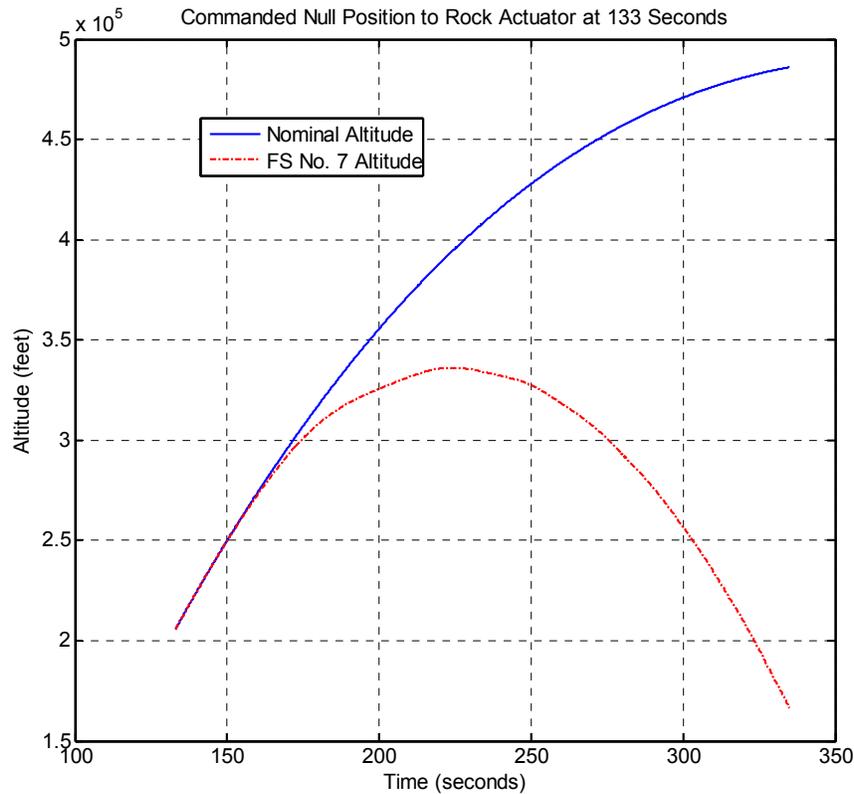


Figure 21.—Comparing altitude responses for nominal and failure scenario seven.

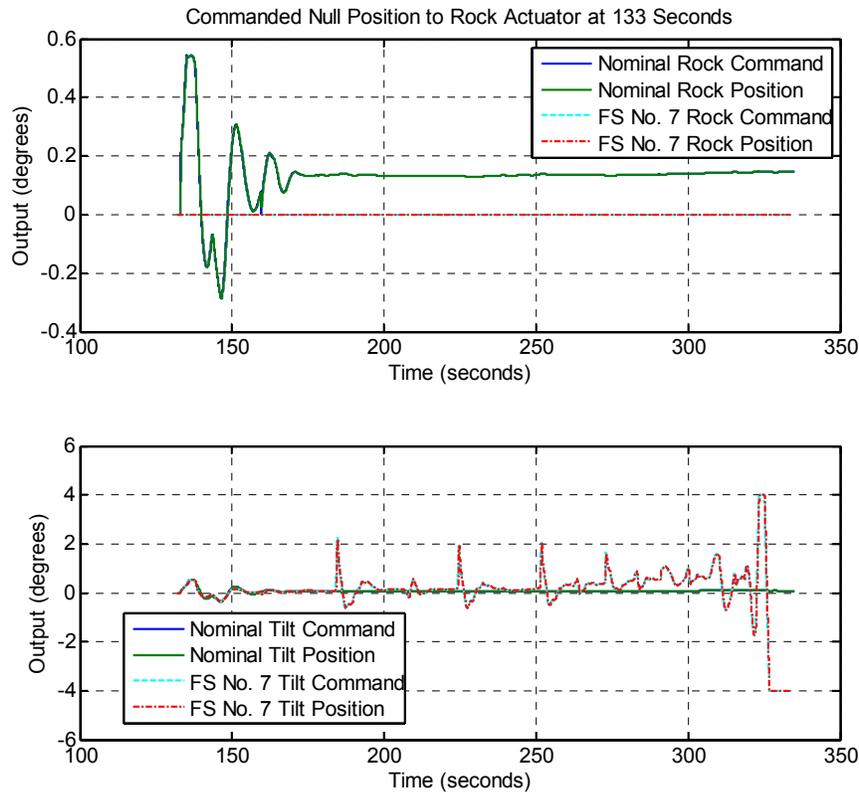


Figure 22.—Comparing output responses for nominal and failure scenario seven.

5.0 Summary and Conclusions

Modeling and simulation provide a means to carry out not only performance analyses that help to define the system architecture, but they also provide the needed resources to investigate off-nominal conditions early in the design phase. This facilitates the development of the software that implements the control systems and diagnostics used by the vehicle. Although, it can be argued that a system's architecture can change significantly during the early stages of a program, it is usually more difficult and expensive to change hardware late in the production phase or afterwards when the vehicle is already in use. It is expected that an investment in modeling, simulation and analysis tools early in the program will provide benefits that include improved safety margins, greater system reliability and reduced maintenance costs. In addition, a hardware-based test program would be expensive to build and implement; therefore, the virtual test bed environment that these models and simulations provide is ideal for the development of controls and diagnostics software.

The simulation results for nominal flight and several failure scenarios have been presented for the Ares I US TVC subsystem. These were produced by a model that integrated two simulation models that represent the US TVC subsystem and the Ares I vehicle. There were a total of seven failure scenarios for which simulation results were shown. Failure scenarios were implemented to ensure the worst possible effect on the vehicle trajectory. Hence, failures were inserted at the beginning of the mission profile at 133 sec when the US TVC subsystem is required to deliver most of its thrust vectoring control. Other scenarios were investigated, which varied the fault insertion time. However, the general response of the system was similar for each type of failure. For example, in case of a hard over condition, the system was unable to meet the mission trajectory even if the failure condition was inserted at the end of the mission profile.

The closed-loop simulation was built in MATLAB Simulink, which contributed to the ease of modeling and testing the failure scenarios. The major components of the US TVC subsystem are accessible for modeling failure scenarios that otherwise would not be possible with a model that does not have the same fidelity. In this case, the original MAVERIC simulation does have a model for the US TVC, but it is not as detailed as the one used here. In order to predict how US TVC failures might impact the mission, a model that is based on system requirements, such as the one used here, might provide more confidence in the conclusions drawn from the simulation results.

During preliminary discussions with simulation designers, who are themselves MAVERIC users, it was suggested we use an integration step size of 0.01 sec for all simulations using MAVERIC. As mentioned before, the Simulink US TVC Actuator subsystem model required the integration step size be variable and could vary to several orders of magnitude below 0.01 sec. During the review of the MAVERIC integrated simulation results, the disparity between the two integration step sizes was questioned. While not affecting the results in this paper, previous modeling experience suggests that the MAVERIC integration step size should be reduced to at least 0.001 sec to more closely match the step size used by the Simulink US TVC subsystem simulation. Engineers, using MAVERIC for Ares I abort condition analysis, used an integration step size of 0.005 sec, so reducing the step size to 0.001 sec would be reasonable.

The MAVERIC simulation breaks down a simulation into calls to various launch vehicle related modules such as the GN&C system, the Flight Control System, Propellant Utilization, the Actuators or the Reaction Control System. MAVERIC also allows users to code their own versions of these modules and incorporate them as plugins. In retrospect, this is the integration path that could have been employed since it would have:

- Aligned with a ‘top down’ design of the integration, i.e., the US TVC subsystem model would have been a subset of models available to MAVERIC.
- Eliminated issues compiling and linking MAVERIC source into the S-function.
- Allowed easier integration with later versions of MAVERIC.

MAVERIC is such a complex simulation that a longer, more detailed review of its capabilities should have been made. For example, MAVERIC has several input files with many input variables, some of which can affect or override each other. While the scope of this task did not call for developing an expertise in MAVERIC, a better knowledge of its capabilities and input structure would have been beneficial during the execution of simulations, which were made for comparing failure scenario results from the MAVERIC standalone simulation and the integrated model.

References

1. Best, Joel; Chavers, Greg; Richardson, Lea; Cruzen, Craig, “Operational Concept for the NASA Constellation Program’s Ares I Crew Launch Vehicle,” SpaceOps 2008 Conference, AIAA-2008-3562, May 12–16, 2008.
2. Davis, Daniel J.; “NASA Ares I Crew Launch Vehicle Upper Stage Overview,” International Astronautical Congress Meeting, IAC-08D.2.3.4, September 28—October 3, 2008.
3. “Thrust Vector Control (TVC) Subsystem Requirements Document (SRD),” GRC-TVC-REQ-001, Revision A, March 30, 2009.
4. “Thrust Vector Control Operations Concept Document,” GRC-TVC-OPS-001, Revision A, September 8, 2009.
5. Connolly, Joseph; Hunker, Keith; Chicatelli, Amy; “Dynamic Modeling of the Ares I Upper Stage Thrust Vector Control for Investigation of Abort Condition Scenarios,” Joint Army-Navy-NASA-Air Force Interagency Propulsion (JANNAF) Meeting December 8–12, 2008.

6. Dynamics Concepts, Inc., “MAVERIC Version 3.1 User’s Guide A Software Tool for the Rapid Development of High-Fidelity 3-DOF/6-DOF Flight Simulations of Aerospace Vehicles,” MSFC contract No. H-36085D, DCI Report TR-FRL-111505-1, August 25, 2005.
7. Hall, Charlie; Lee, Chong; Jackson, Mark; Whorton, Mark; West, Mark; Brandon, Jay; Hall, Rob A.; Jang, Jiann-Woei; Bedrossian, Naz; Compton, Jimmy; Rutherford, Chad; “Ares I Flight Control System Overview,” AIAA Guidance, Navigation and Control Conference, Honolulu, HI, August 18–21, 2008.
8. Hanson, John M.; Hall, Charlie E., “Learning about Ares I from Monte Carlo Simulation,” 2008 AIAA Guidance, Navigation and Control Conference, Honolulu, HI, August 18–21, 2008.

REPORT DOCUMENTATION PAGE			Form Approved OMB No. 0704-0188		
<p>The public reporting burden for this collection of information is estimated to average 1 hour per response, including the time for reviewing instructions, searching existing data sources, gathering and maintaining the data needed, and completing and reviewing the collection of information. Send comments regarding this burden estimate or any other aspect of this collection of information, including suggestions for reducing this burden, to Department of Defense, Washington Headquarters Services, Directorate for Information Operations and Reports (0704-0188), 1215 Jefferson Davis Highway, Suite 1204, Arlington, VA 22202-4302. Respondents should be aware that notwithstanding any other provision of law, no person shall be subject to any penalty for failing to comply with a collection of information if it does not display a currently valid OMB control number.</p> <p>PLEASE DO NOT RETURN YOUR FORM TO THE ABOVE ADDRESS.</p>					
1. REPORT DATE (DD-MM-YYYY) 01-08-2010		2. REPORT TYPE Technical Memorandum		3. DATES COVERED (From - To)	
4. TITLE AND SUBTITLE Closed-Loop Simulation Study of the Ares I Upper Stage Thrust Vector Control Subsystem for Nominal and Failure Scenarios			5a. CONTRACT NUMBER		
			5b. GRANT NUMBER		
			5c. PROGRAM ELEMENT NUMBER		
6. AUTHOR(S) Chicattelli, Amy; Fulton, Chris; Connolly, Joe; Hunker, Keith			5d. PROJECT NUMBER		
			5e. TASK NUMBER		
			5f. WORK UNIT NUMBER WBS 136905.08.05.07.01.03		
7. PERFORMING ORGANIZATION NAME(S) AND ADDRESS(ES) National Aeronautics and Space Administration John H. Glenn Research Center at Lewis Field Cleveland, Ohio 44135-3191			8. PERFORMING ORGANIZATION REPORT NUMBER E-17329		
9. SPONSORING/MONITORING AGENCY NAME(S) AND ADDRESS(ES) National Aeronautics and Space Administration Washington, DC 20546-0001			10. SPONSORING/MONITOR'S ACRONYM(S) NASA		
			11. SPONSORING/MONITORING REPORT NUMBER NASA/TM-2010-216753		
12. DISTRIBUTION/AVAILABILITY STATEMENT Unclassified-Unlimited Subject Category: 15 Available electronically at http://gltrs.grc.nasa.gov This publication is available from the NASA Center for AeroSpace Information, 443-757-5802					
13. SUPPLEMENTARY NOTES					
14. ABSTRACT As a replacement to the current Shuttle, the Ares I rocket and Orion crew module are currently under development by the National Aeronautics and Space Administration (NASA). This new launch vehicle is segmented into major elements, one of which is the Upper Stage (US). The US is further broken down into subsystems, one of which is the Thrust Vector Control (TVC) subsystem which gimbals the US rocket nozzle. Nominal and off-nominal simulations for the US TVC subsystem are needed in order to support the development of software used for control systems and diagnostics. In addition, a clear and complete understanding of the effect of off-nominal conditions on the vehicle flight dynamics is desired. To achieve these goals, a simulation of the US TVC subsystem combined with the Ares I vehicle as developed. This closed-loop dynamic model was created using Matlab's Simulink and a modified version of a vehicle simulation, MAVERIC, which is currently used in the Ares I project and was developed by the Marshall Space Flight Center (MSFC). For this report, the effects on the flight trajectory of the Ares I vehicle are investigated after failures are injected into the US TVC subsystem. The comparisons of the off-nominal conditions observed in the US TVC subsystem with those of the Ares I vehicle flight dynamics are of particular interest.					
15. SUBJECT TERMS Launch vehicles					
16. SECURITY CLASSIFICATION OF:			17. LIMITATION OF ABSTRACT UU	18. NUMBER OF PAGES 32	19a. NAME OF RESPONSIBLE PERSON STI Help Desk (email:help@sti.nasa.gov)
a. REPORT U	b. ABSTRACT U	c. THIS PAGE U			19b. TELEPHONE NUMBER (include area code) 443-757-5802

